# Chapter 6

# Basic flows

## 6.1 Uniform Flow at An Angle $\alpha$

Given velocity field is:

$$\vec{V} = (V_\infty \cos\alpha, V_\infty \sin\alpha)$$

Check if conservation of mass is satisfied first to test if it is a physically possible flow?

$$\nabla \cdot \vec{V} \stackrel{?}{=} 0$$
$$\frac{\partial u}{\partial x} + \frac{\partial v}{\partial y} \stackrel{?}{=} 0$$

Since $u$ and $v$ are both constants, $\nabla \cdot \vec{V} = 0$
Therefore $\psi$ exists. From conservation of mass,

$$u = \frac{\partial \psi}{\partial y} \quad \text{and} \quad v = -\frac{\partial \psi}{\partial x}$$
$$u = V_\infty \cos\alpha = \frac{\partial \psi}{\partial y}$$
$$= V_\infty \cos\alpha \;\; y + f(x)$$
$$\frac{\partial \psi}{\partial x} = -v = 0 + f'(x)$$
$$f'(x) = -V_\infty \sin\alpha \quad or \quad f(x) = -V_\infty \sin\alpha \;\; x + g(y)$$
$$= -V_\infty \sin\alpha \;\; x + V_\infty \cos\alpha \;\; y$$
$$= const. = -V_\infty \sin\alpha \;\; x + V_\infty \cos\alpha \;\; y$$
$$\frac{}{V_\infty} = -\sin\alpha \;\; x + \cos\alpha \;\; y$$
$$\frac{}{V_\infty \cos\alpha} = -\tan\alpha \;\; x + y$$

Equation of streamlines:

$$y = tan\alpha \;\; x + \frac{}{V_\infty \cos\alpha}$$

Check if the given flow is a potential flow?

$$\nabla \times \vec{V} \stackrel{?}{=} 0$$
$$\frac{\partial v}{\partial x} - \frac{\partial u}{\partial y} \stackrel{?}{=} 0$$

Since $V_\infty$ and $\alpha$ are constant throughtout the flow, $\nabla \times \vec{V} = 0$
Therefore $\phi$ exists and $\vec{V} = \nabla\phi$.

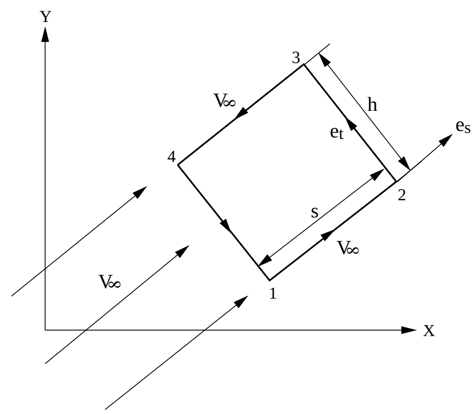
$$\vec{V} = \nabla\phi = \frac{\partial\phi}{\partial x}\hat{\imath} + \frac{\partial\phi}{\partial y}\hat{\jmath}$$
$$\frac{\partial\phi}{\partial x} = V_\infty \cos\alpha = u$$
$$\phi = V_\infty \cos\alpha \;\; x + f(y)$$
$$\frac{\partial\phi}{\partial y} = 0 + f'(y) = v$$
$$f'(y) = V_\infty \sin\alpha \quad or \quad f(y) = V_\infty \sin\alpha \;\; y + f(x)$$
$$\phi = V_\infty \cos\alpha \;\; x + V_\infty \sin\alpha \;\; y \quad \text{(uniform flow at an angle } \alpha)$$
$$\phi = const. = V_\infty \cos\alpha \;\; x + V_\infty \sin\alpha \;\; y$$
$$\frac{\phi}{V_\infty \sin\alpha} = \frac{x}{\tan\alpha} + y$$

Equation of Equipotential lines:

$$y = -\frac{1}{\tan\alpha}x + \frac{\phi}{V_\infty \sin\alpha}$$

$\phi$ constant lines are orthogonal to $\psi$ constant lines.

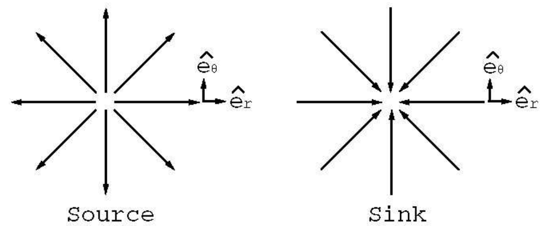
### 6.1.1 $\Gamma$: Contour Integral over a Close Curve$C$




$$\Gamma = -\oint \vec{V} \cdot d\vec{l}$$
$$= -\left[\int_1^2 \vec{V} \cdot d\vec{l} + \int_2^3 \vec{V} \cdot d\vec{l} + \int_3^4 \vec{V} \cdot d\vec{l} + \int_4^1 \vec{V} \cdot d\vec{l}\right]$$
$$= -\left[\int_1^2 (V_\infty \hat{e}_s) \cdot (ds\ \hat{e}_s) + \int_2^3 (V_\infty \hat{e}_s) \cdot (dh\ \hat{e}_t) + \int_3^4 (V_\infty \hat{e}_s) \cdot (-ds\ \hat{e}_s) + \int_4^1 (V_\infty \hat{e}_s) \cdot (-dh\ \hat{e}_t)\right]$$
$$= -\left[(V_\infty s) + 0 + (-V_\infty s) + 0\right] \equiv 0$$

## 6.2 2-D Source (Line Source)

Definition: A source is a point from which fluid issues along radial lines. Streamlines are straight lines emanating from a central point. Velocity varies inversely with distance from the origin.




From the definition of the source the velocity vector can be written as:

$$\vec{V} = v_r \hat{e}_r$$

where $v_r \propto \frac{1}{r}$ or $v_r = \frac{c}{r}$, and $v_\theta = 0$ where $C$ is a constant.
Check if the assumed flow is physically possible.

$$\vec{V} = \frac{c}{r}\hat{e}_r + 0\hat{e}_\theta$$

$$\nabla \cdot \vec{V} \stackrel{?}{=} 0$$

$$\nabla \cdot \vec{V} = \frac{1}{r}\left[\frac{\partial(rv_r)}{\partial r} + \frac{\partial(v_\theta)}{\partial \theta}\right] = \frac{1}{r}\left[\frac{\partial(c)}{\partial r} + \frac{\partial(0)}{\partial \theta}\right] \equiv 0$$

Flow is physically possible and $\psi$ exists.

### 6.2.1 Evaluation of $c$

From mass conservation for a steady flow we know

$$\int d\dot{m} = 0$$

From continuity the mass of fluid per unit time crossing any circle centered at the source is a constant and equal to the mass of fluid issuing per unit time from the source. Consider a cylinder centered on the source. There is mass flowing out only from the sides of the cylinder.

$$d\vec{A}_r = h_\theta h_z d\theta \; dz\hat{e}_r = r \; d\theta \; dz\hat{e}_r$$

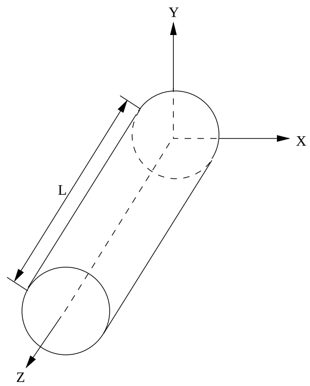
$$\dot{m} = \int_0^L \int_0^{2\pi} \rho\vec{V} \cdot d\vec{A} = \int_0^L \int_0^{2\pi} \rho(V_r \; \hat{e}_r) \cdot (r \; d\theta \; dz)\hat{e}_r$$

It is a 2-D flow and hence the integral can be reduced to:

$$\dot{m} = L\int_0^{2\pi} \rho V_r \; r \; d\theta$$

$V_r$ is not a function of $\theta$. $V_r$ is only a function of $r$.

$$\dot{m} = L\int_0^{2\pi} \rho\left(\frac{c}{r}\right) r \; d\theta = \rho \; L \; c \; 2\pi$$




Cylinder centered at a source

Volume flow per second is:

$$\frac{\dot{m}}{\rho} = c\ 2\pi\ L$$

Define $K$ as the source strength. It is physically the rate of volume flow from the source per unit depth into the page (2-D).

$$K = 2\pi c \quad or \quad c = \frac{K}{2\pi}$$

then the velocity becomes:

$$v_r = \frac{K}{2\pi r}$$

Since $\nabla \cdot \vec{V} = 0$ is satisfied, the flow is physically possible and from the definition of $\psi$ in polar coordinates, can be found.

$$v_r = \frac{\partial\psi}{r\partial\theta} \quad and \quad v_\theta = -\frac{\partial\psi}{\partial r}$$

$$\frac{\partial\psi}{r\partial\theta} = \frac{K}{2\pi r}$$

$$\frac{\partial\psi}{\partial\theta} = \frac{K}{2\pi}$$

$$= \frac{K}{2\pi}\theta + f(r)$$

$$\frac{\partial\psi}{\partial r} = -v_\theta = 0 + f'(r) = 0$$

$$f(r) = const.$$

$$= \frac{K}{2\pi}\theta$$

Since the source strength, $K$ is a constant, $\psi$ constant lines are radial lines.

$$= \frac{K}{2\pi}\theta = const.$$

Is $\nabla \times \vec{V} \stackrel{?}{=} 0$.

$$\nabla \times \vec{V} = \frac{1}{r}\begin{vmatrix} \hat{e}_r & r\hat{e}_\theta & \hat{e}_z \\ \frac{\partial}{\partial r} & \frac{\partial}{\partial\theta} & \frac{\partial}{\partial z} \\ v_r & rv_\theta & v_z \end{vmatrix} = \frac{1}{r}\left[\frac{\partial(rv_\theta)}{\partial r} - \frac{\partial(v_r)}{\partial\theta}\right]\hat{e}_z \equiv 0$$

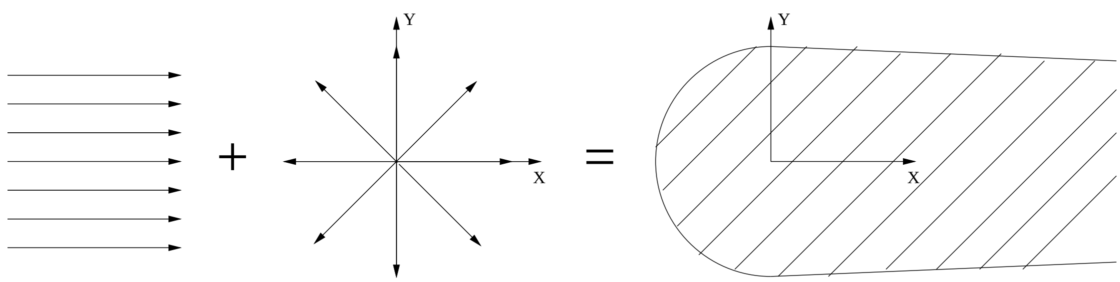
therefore $\phi$ exists.

$$
\begin{aligned}
& v_r = \frac{\partial \phi}{\partial r} = \frac{K}{2\pi r} \quad and \quad v_\theta = \frac{1}{r}\frac{\partial \phi}{\partial \theta} = 0 \\
& \phi = \frac{K}{2\pi}\ln r + g(\theta) \\
& \frac{\partial \phi}{\partial \theta} = 0 + g'(\theta) = 0 \\
& g(\theta) = const. \\
& \therefore \phi = \frac{K}{2\pi}\ln r
\end{aligned}
$$

## 6.3 Combination of Potential Flows

Uniform flow and source/sink satisfy Laplace equation and therefore superposition is possible.

### 6.3.1 Combination of a Uniform Flow to The Right ($\alpha = 0$) and A Source at The Origin




| Quantity | Uniform flow | Source/Sink | Combination |
|---|---|---|---|
| $\vec{V}$ | $V_\infty \hat{\imath}$ | $\pm\frac{K}{2\pi r}\hat{e}_r$ | $V_\infty \hat{\imath} \pm \frac{K}{2\pi r}\hat{e}_r$ |
| $\phi$ | $V_\infty x$ | $\pm\frac{K}{2\pi}\ln r$ | $V_\infty x \pm \frac{K}{2\pi}\ln r$ |
| | $V_\infty y$ | $\pm\frac{K}{2\pi}\theta$ | $V_\infty y \pm \frac{K}{2\pi}\theta$ |

**Stagnation Point:**

At the stagnation point, $\vec{V} \equiv 0$

$$
\begin{aligned}
& v_r = V_\infty \cos\theta + \frac{K}{2\pi r} = 0 \\
& v_\theta = -V_\infty \sin\theta = 0
\end{aligned}
$$

Solve for $\theta$ and $r$ at the stagnation point to get $(r_{stag}, \theta_{stag})$. Proceed to find $\psi_{stag}$ to get the shape of the body. From $v_\theta = 0$:

$$
\begin{aligned}
& \sin\theta = 0 \\
& \theta = 0 \;\; or \;\; \pm\pi
\end{aligned}
$$

*Case 1:* $\theta_s = 0$. Sove for $r_s$ from $v_r$=0

$$\begin{aligned} v_r &= V_\infty \cos\theta + \frac{K}{2\pi r} = 0 \\ \text{if} \quad \theta &= \theta_s = 0 \\ \cos\theta_s &= 1 \\ v_r &= V_\infty + \frac{K}{2\pi r_s} = 0 \\ \text{or} \quad r_s &= -\frac{K}{2\pi V_\infty} \end{aligned}$$

Impossible solution as $r_s < 0$; ($\because K$ and $V_\infty$ are positive)
*Case 2:* $\theta_s = \pm\pi$.

$$\begin{aligned} v_r &= -V_\infty + \frac{K}{2\pi r_s} = 0 \\ r_s &= \frac{K}{2\pi V_\infty} \end{aligned}$$

$\theta_s = +\pi$ for the upper half of the body
$\theta_s = -\pi$ for the lower half of the body
Coordinates of the stagnation point:

$$(r_s, \theta_s) = \left(\frac{K}{2\pi V_\infty}, \pm\pi\right)$$

**Body Shape (Stagnation Streamline):**

A general expression for the streamfunction for the combined flow is:

$$= V_\infty r \sin\theta + \frac{K}{2\pi}\theta$$

Find $\psi_s$ (= the body shape) by substituting ($r_s$, $\theta_s$).

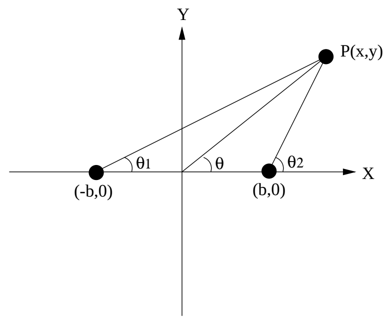
$$\psi_s = V_\infty r \sin(\pm\pi) + \frac{K}{2\pi}(\pm\pi) = \pm\frac{K}{2} = const.$$

In Cartesian coordinate, the general expression for the body streamfunction becomes:

$$\begin{aligned} \pm\frac{K}{2} &= V_\infty y + \frac{K}{2\pi}\tan^{-1}\left(\frac{y}{x}\right) \\ \frac{K}{2\pi}\tan^{-1}\left(\frac{y}{x}\right) &= \left(\pm\frac{K}{2} - V_\infty y\right) \\ \tan^{-1}\left(\frac{y}{x}\right) &= \left(\pm\pi - \frac{2\pi V_\infty y}{K}\right) \\ \frac{y}{x} &= \tan\left(\pm\pi - \frac{2\pi V_\infty y}{K}\right) \\ x &= \frac{y}{\tan\left(\pm\pi - \frac{2\pi V_\infty y}{K}\right)} \end{aligned}$$

To find maximum $y$ value, consider $\psi = K/2$ (upper half of the body).

$$\begin{aligned} \frac{K}{2} &= V_\infty y + \frac{K}{2\pi}\tan^{-1}\left(\frac{y}{x}\right) \\ y_{max} &= y_{@x=\infty} = \frac{K}{2V_\infty} \end{aligned}$$




### 6.3.2 Combined Flow of a Source at $(-b,0)$ and a Sink at $(b,0)$

$$= \frac{K}{2\pi}\theta_1 - \frac{K}{2\pi}\theta_2$$

where $\theta_1$ and $\theta_2$ are measured from the center of the source and sink respectively.

$$\theta_1 = \tan^{-1}\left(\frac{y}{x+b}\right), \quad \theta_2 = \tan^{-1}\left(\frac{y}{x-b}\right)$$

$$\theta_2 - \theta_1 = \tan^{-1}\left(\frac{y}{x-b}\right) - \tan^{-1}\left(\frac{y}{x+b}\right)$$

$$\theta_2 - \theta_1 = \tan^{-1}\left(\frac{2by}{x^2+y^2-b^2}\right)$$

$$= \psi_1 + \psi_2 = \frac{K}{2\pi}(\theta_1 - \theta_2)$$

$$\theta_1 - \theta_2 = -\tan^{-1}\left(\frac{2by}{x^2+y^2-b^2}\right)$$

$$\psi_{source+sink} = -\frac{K}{2\pi}\tan^{-1}\left(\frac{2by}{x^2+y^2-b^2}\right)$$

$$\frac{2\pi\psi}{K} = -\tan^{-1}\left(\frac{2by}{x^2+y^2-b^2}\right)$$

$$\tan\left(\frac{2\pi}{K}\right) = -\frac{2by}{x^2+y^2-b^2}$$

$$x^2 + y^2 + 2by\cot\left(\frac{2\pi}{K}\right) = b^2$$

$$(x-0)^2 + \left(y + b\cot\left[\frac{2\pi}{K}\right]\right)^2 = b^2\left(1 + \cot^2\left[\frac{2\pi}{K}\right]\right)$$

$$(x-0)^2 + \left(y + b\cot\left[\frac{2\pi}{K}\right]\right)^2 = b^2\csc^2\left[\frac{2\pi}{K}\right]$$

Equation of a circle with center at $\left(0, \pm b\cot\frac{2\pi\psi}{K}\right)$ and radius of $\left(b\csc\frac{2\pi\psi}{K}\right)$. When $y = 0$, $x = \pm b$. All streamlines go through $\pm b$.

### 6.3.3 Uniform Flow to The Right+Source $(-b,0)$+ Sink $(b,0)$ (Rankine oval)

- Source of strength $K$ placed at $(-b,0)$
- Sink of strength $K$ placed at $(b,0)$
- Uniform flow to the right $(\alpha = 0)$

$$= V_\infty r \sin\theta + \frac{K}{2\pi}\theta_1 - \frac{K}{2\pi}\theta_2$$

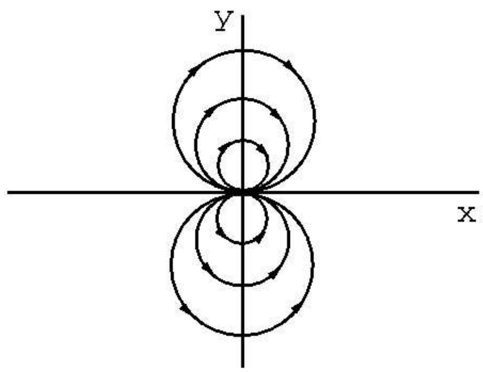
**Problem:**

Analyze Rankine oval.

### 6.3.4 2-D Doublet

Definition: A doublet is obtained when a source and sink of equal strength approach each other so that the product of their strength and the distance apart remains a constant.

$$B = K(2b) = constant$$




Doublet

$$\psi_{source+sink} = -\frac{K}{2\pi}\tan^{-1}\left(\frac{2by}{x^2+y^2-b^2}\right)$$

$$= -\frac{2bK}{4\pi}\frac{\tan^{-1}\left(\frac{2by}{x^2+y^2-b^2}\right)}{b}$$

$$\lim_{2bK\to\mu}\psi_{source+sink} = \psi_{doublet} = -\frac{\mu}{4\pi}\lim_{b\to 0}\left[\frac{\tan^{-1}\left(\frac{2by}{x^2+y^2-b^2}\right)}{b}\right]$$

Using L'Hospital's rule:

$$\psi_{doublet} = -\frac{\mu}{4\pi}\lim_{b\to 0}\left[\frac{\frac{\frac{d}{db}\frac{2by}{x^2+y^2-b^2}}{1+\left(\frac{2by}{x^2+y^2-b^2}\right)^2}}{\frac{db}{db}}\right] = -\frac{\mu}{4\pi}\lim_{b\to 0}\left[\frac{\frac{(x^2+y^2-b^2)(2y)-(2by)(-2b)}{(x^2+y^2-b^2)^2}}{1+\left(\frac{2by}{x^2+y^2-b^2}\right)^2}\right] = -\frac{\mu}{4\pi}\left(\frac{2y}{x^2+y^2}\right)$$

$$\psi_{doublet} = -\frac{\mu}{2\pi}\frac{\sin\theta}{r}$$

**Streamlines**

$$x^2+y^2+\frac{\mu y}{2\pi\psi} = 0$$

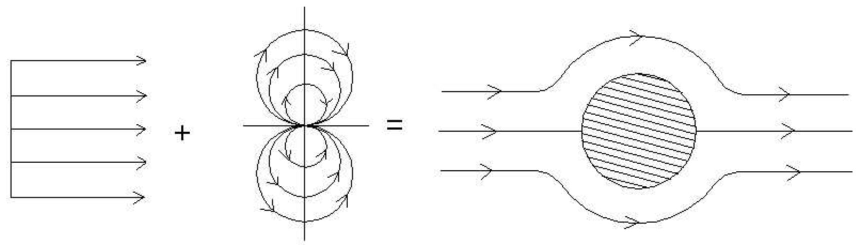
$$(x-0)^2 + \left(y+\frac{\mu}{4\pi\psi}\right)^2 = \left(\frac{\mu}{4\pi\psi}\right)^2$$

Streamlines are circles centerd on the $y$-axis a distance $-\frac{\mu}{4\pi\psi}$ from the $x$-axis with a radius of $\left|\frac{\mu}{4\pi\psi}\right|$. All circles pass through the origin.

**Problem:**

Show that $\phi = \frac{\mu}{2\pi}\frac{\cos\theta}{r}$ for a 2-D doublet.

### 6.3.5 Uniform Flow to The Right + A 2-D Doublet




| Quantity | Uniform flow | 2-D doublet | Combination |
|---|---|---|---|
| $\vec{V}$ | $V_\infty \hat{\imath}$ | | |
| $\phi$ | $V_\infty x$ | $\frac{\mu}{2\pi}\frac{\cos\theta}{r}$ | $V_\infty x + \frac{\mu}{2\pi}\frac{\cos\theta}{r}$ |
| | $V_\infty y$ | $-\frac{\mu}{2\pi}\frac{\sin\theta}{r}$ | $V_\infty y - \frac{\mu}{2\pi}\frac{\sin\theta}{r}$ |

$$V_r = \frac{1}{r}\frac{\partial\psi}{\partial\theta} = \frac{1}{r}\left[V_\infty r\cos\theta - \frac{\mu}{2\pi}\frac{\cos\theta}{r}\right]$$

$$= V_\infty\cos\theta\left[1 - \underbrace{\frac{\mu}{2\pi V_\infty}}_{1/R^2}\frac{1}{r^2}\right] = V_\infty\cos\theta\left[1 - \left(\frac{R}{r}\right)^2\right]$$

$$V_\theta = -\frac{\partial\psi}{\partial r} = -\left[V_\infty + \frac{\mu}{2\pi r^2}\right]\sin\theta = -V_\infty\sin\theta\left[1 + \left(\frac{R}{r}\right)^2\right]$$

Where $R^2 = \frac{\mu}{2\pi V_\infty}$.

**Stagnation Points ($\vec{V} = 0$)**

Set $V_\theta = 0$.

$$0 = -V_\infty\sin\theta\left[1 + \left(\frac{R}{r}\right)^2\right]$$

$$\sin\theta = 0 \quad \text{or} \quad \theta_s = (0 \ \text{ or } \ \pi)$$

Now set $V_r = 0$.

$$0 = V_\infty\sin\theta\left[1 - \left(\frac{R}{r}\right)^2\right]$$

$$\text{For } \theta = 0 \ \text{ or } \ \pi, \quad \cos\theta \neq 0$$

$$\therefore \left[1 - \left(\frac{R}{r}\right)^2\right] \equiv 0 \ \text{ or } \ r^2 = R^2 = \frac{\mu}{2\pi V_\infty}$$

The stagnation points are located at

$$(r_s, \theta_s) \equiv (R, 0) \text{ and } (R, \pi)$$

**For Cylindrical System**

$$\left.\begin{aligned} \phi &= V_\infty r \cos\theta \left(1 + \frac{R^2}{r^2}\right) \\ &= V_\infty r \sin\theta \left(1 - \frac{R^2}{r^2}\right) \\ V_r &= V_\infty \cos\theta \left(1 - \frac{R^2}{r^2}\right) \\ V_\theta &= -V_\infty \sin\theta \left(1 + \frac{R^2}{r^2}\right) \end{aligned}\right\} (r \geq R)$$

Where $R^2 = \dfrac{\mu}{2\pi V_\infty}$

Substitute $(r_s, \theta_s) = (R, 0)$ or $(R, \pi)$ in the expression for $\psi$.

$$\begin{aligned} &\psi_s = 0 \\ &\text{at } r = R \text{ (surface of the cylinder)} \\ &V_r = V_\infty \cos\theta \left(1 - \frac{R^2}{r^2}\right) = 0 \text{ (no flow out of the cylinder)} \\ &V_\theta = -2V_\infty \sin\theta \\ &C_p|_{r=R} = \frac{p - p_\infty}{\frac{1}{2}\rho_\infty V_\infty} = 1 - \left(\frac{V}{V_\infty}\right)^2 = 1 - \frac{V_r^2 + V_\theta^2}{V\infty^2} = 1 - \left(\frac{V_\theta}{V_\infty}\right)^2 = 1 - 4\sin^2\theta \\ &C_p \text{ (2-D cylinder) } = 1 - 4\sin^2\theta \end{aligned}$$

## 6.4 2-D Vortex Flow (Potential Vortex)

A 2-D point vortex is a mathematical concept that induces a velocity field given by

$$V_r = 0, \quad V_\theta = \frac{const.}{r} = \frac{C}{r}$$

1. Check if the flow satisfies conservation of mass (Is it a physically possible flow?)

$$\begin{aligned} &\nabla \cdot \vec{V} \stackrel{?}{=} 0 \\ &\nabla \cdot \vec{V} = \frac{1}{r}\left[\frac{\partial(V_r r)}{\partial r} + \frac{\partial V_\theta}{\partial \theta}\right] = 0 \rightarrow \quad \text{exist.} \\ &V_r = \frac{\partial \psi}{r\partial\theta} = 0 \rightarrow \psi = g(r) \\ &V_\theta = -\frac{\partial\psi}{\partial r} = \frac{C}{r} \rightarrow \quad = -C\ln r + f(\theta) \\ &\frac{\partial\psi}{\partial\theta} = f'(\theta) = 0 \\ &f(\theta) = const. \\ &\therefore \psi = -C\ln r + const. \end{aligned}$$

When $r \to 0$, $V_\theta = \infty$ and $\quad \to \infty$. To eliminate the infinite velocity it is arbitrary assumed that $\quad = 0$ at $r = R$

$$\begin{aligned} \therefore \psi &= -C \ln R + const. = 0 \\ const. &= C \ln R \\ &= -C \ln\left(\frac{r}{R}\right) \quad \text{for} \quad (r \geq R) \end{aligned}$$

2. Check if the flow is irrotational

$$\nabla \times \vec{V} \stackrel{?}{=} 0$$

$$\nabla \times \vec{V} = \frac{1}{h_1 h_2 h_3} \begin{vmatrix} h_1\hat{e}_1 & h_2\hat{e}_2 & h_3\hat{e}_3 \\ \frac{\partial}{\partial q_1} & \frac{\partial}{\partial q_2} & \frac{\partial}{\partial q_3} \\ h_1 V_1 & h_2 V_2 & h_3 V_3 \end{vmatrix} = \frac{1}{r} \begin{vmatrix} \hat{e}_r & r\hat{e}_\theta & \hat{e}_z \\ \frac{\partial}{\partial r} & \frac{\partial}{\partial \theta} & \frac{\partial}{\partial z} \\ V_r & rV_\theta & V_z \end{vmatrix}$$

Vorticity or $(\nabla \times \vec{V})$ in the $r - \theta$ plane

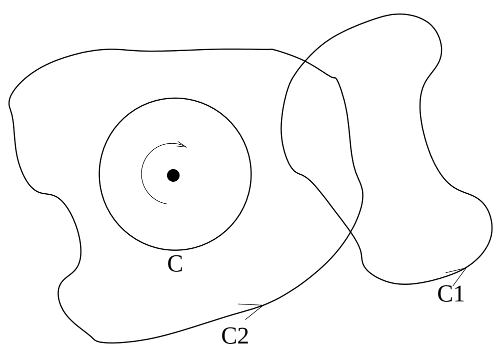
$$\frac{1}{r}\left(\frac{\partial r V_\theta}{\partial r} - \frac{\partial V_r}{\partial \theta}\right) = \left(\frac{\partial C}{\partial r} - \frac{\partial 0}{\partial \theta}\right) = 0 \to \phi \text{ exist.}$$

**Problem:**

Show that $\phi = \quad C\theta$.

**Evaluate the Constant $C$**

Evaluate the circulation $\Gamma$ around the point vortex.




1. Around closed curve $C1$ that does not include the point vortex

$$\Gamma_{C1} = -\oint_{C1} \vec{V} \cdot d\vec{l} = \iint_{S_1} (\underbrace{\nabla \times \vec{V}}_{0}) \cdot d\vec{A} = 0$$

2. Around $C2$ that includes the point vortex.

$$\Gamma_{C2} = -\left[\oint_{C2} (V_r\hat{e}_r + V_\theta\hat{e}_\theta)\cdot(dr\ \hat{e}_r + r\ d\theta\hat{e}_\theta)\right]$$

$$= -\left[\oint_{C} (V_r\hat{e}_r + V_\theta\hat{e}_\theta)\cdot(dr\ \hat{e}_r + r\ d\theta\hat{e}_\theta)\right] + \left[\oint_{C2-C} (V_r\hat{e}_r + V_\theta\hat{e}_\theta)\cdot(dr\ \hat{e}_r + r\ d\theta\hat{e}_\theta)\right]$$

$$= -\left[\oint_{C} (V_r\hat{e}_r + V_\theta\hat{e}_\theta)\cdot(dr\ \hat{e}_r + r\ d\theta\hat{e}_\theta) + 0\right]$$

$$= -\left[\oint_{C} V_r dr + \oint_{C} V_\theta r\ d\theta\right] = -\left[0 + \int_0^{2\pi}\left(\frac{C}{r}\right) r\ d\theta\right] = -2\pi C$$

$$\Gamma_{C2} = -2\pi C \text{ or } -\frac{\Gamma}{2\pi} = C$$

This implies that the circulation evaluated for a curve enclosing the 2-D vortex is a constant and not equal to zero.

For a potential vortex, $V_\theta = -\frac{\Gamma}{2\pi r}$ and $\psi = -C\ln\frac{r}{R}$.

$$\therefore \psi = \frac{\Gamma}{2\pi}\ln\frac{r}{R} \text{ for } r \geq R$$

$= const$ , then $\ln\frac{r}{R} = \frac{2\pi\psi}{\Gamma}$, $\frac{r}{R} = e^{2\pi\psi/\Gamma}$, $r = Re^{2\pi\psi/\Gamma}$

Streamlines are concentric circles with center at the 2D point vortex.

$V_r = \frac{\partial\phi}{\partial r}$ , $V_\theta = \frac{1}{r}\frac{\partial\phi}{\partial\theta} = -\frac{\Gamma}{2\pi r}$

$\phi = -\frac{\Gamma}{2\pi}\theta + C$ or $\phi = -\frac{\Gamma}{2\pi}\theta$ (straight lines form the origin).

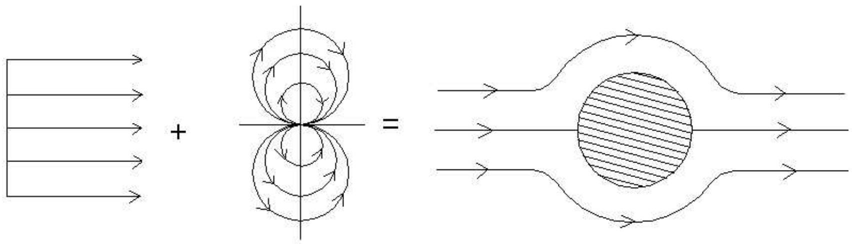
A line vortex can be described as a string of rotating particles. A chain of fluid particles are spinning on their common axis and carrying around with them a swirl of particles which flow around in circles.

A cross-section of such a string of particles and its associated flow shows a spinning point 'outside' of which is streamline flow in concentric circles.

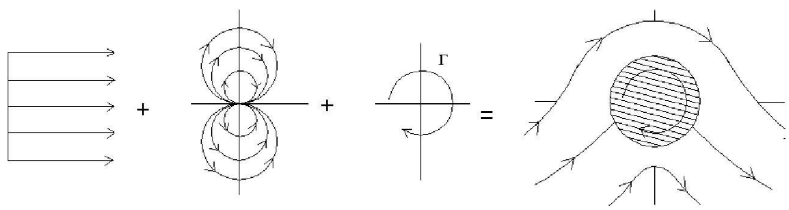
Vortices are common in nature, the difference between a real vortex as opposed to a theoretical line vortex is that the former has a core of fluid which is rotating as a 'solid', although the associated 'swirl' outside is the same as the flow 'outside' the point vortex.

### 6.4.1 Uniform Flow to The Right ($\alpha = 0$) + A 2-D Doublet + A 2-D Point Vortex

- As we all know, uniform flow to the right + 2-D Doublet = non-lifting over a cylinder



- Uniform flow to the right + 2-D Doublet + 2-D Point Vortex = Lifting flow over a cylinder




The parameters for lifting flow over a cylinder are as follow (spinning cylinder):

| Quantity | Non-lifting flow over a cylinder | Vortex of Strength $\Gamma$ | Combination |
|---|---|---|---|
| | $V_\infty r \sin\theta(1 - \frac{R^2}{r^2})$ | $\frac{\Gamma}{2\pi}\ln\frac{r}{R}$ | $V_\infty r \sin\theta(1 - \frac{R^2}{r^2}) + \frac{\Gamma}{2\pi}\ln\frac{r}{R}$ |
| $\phi$ | $V_\infty r \cos\theta(1 + \frac{R^2}{r^2})$ | $-\frac{\Gamma}{2\pi}\theta$ | $V_\infty r \cos\theta(1 + \frac{R^2}{r^2}) - \frac{\Gamma}{2\pi}\theta$ |
| $V_r$ | $V_\infty \cos\theta(1 - \frac{R^2}{r^2})$ | $0$ | $V_\infty \cos\theta(1 - \frac{R^2}{r^2})$ |
| $V_\theta$ | $-V_\infty \sin\theta(1 + \frac{R^2}{r^2})$ | $-\frac{\Gamma}{2\pi r}$ | $-V_\infty \sin\theta(1 + \frac{R^2}{r^2}) - \frac{\Gamma}{2\pi r}$ |

- Flow satisfies continuity at every point $r \geq R$.
  $\therefore \nabla \cdot \vec{V} = 0$.
- Flow satisfies irrotationality at every point $r \geq R$.
  $\therefore \nabla \times \vec{V} = 0$.

**Determine the stagnation points for the combined flow**

At the stagnation points, $\vec{V} = 0$,$V_r = 0 = V_\theta$. If we set $V_r = 0$, we get $r_s = R$ *or* $\theta_s = \pm\frac{\pi}{2}$,

<u>*Case*(1)</u>: $r = R_s = R$

$$V_\theta = -V_\infty \sin\theta_s(1+1) - \frac{\Gamma}{2\pi R} = 0$$

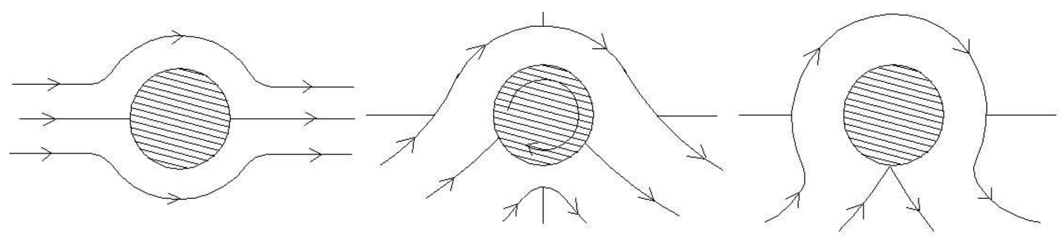
$$sin\theta_s = -\frac{\Gamma}{4\pi R V_\infty} \leq 0$$

Because $\Gamma > 0$, $4\pi R V_\infty > 0$, $\frac{\Gamma}{4\pi V_\infty} > 0$. When $\frac{\Gamma}{4\pi V_\infty} < R$, $\theta_s$ has one value in the third quadrant and one in the fourth quadrant that will satisfy the above relation.
The coordinates of the stagnation point are:

$$y_s = Rsin\theta_s = -\frac{\Gamma}{4\pi V_\infty}$$

$$x_s = \pm\sqrt{R^2 - y_s^2} = \pm\sqrt{R^2 - (\frac{\Gamma}{4\pi V_\infty})^2}$$

When $\frac{\Gamma}{4\pi V_\infty} = R$, there is only one solution. However, the method fails when $\frac{\Gamma}{4\pi V_\infty} > R$.



$\Gamma = 0$ $\quad$ $0 < \frac{\Gamma}{4\pi V_\infty} < R$ $\quad$ $\frac{\Gamma}{4\pi V_\infty} = R$

$\underline{Case(2)}$: $\theta = \pm\frac{\pi}{2}$

$\underline{Case(2a)}$: $\theta = \frac{\pi}{2}$, $r = r_s$

$$V_r = V_\infty \cos(\frac{\pi}{2})(1 - \frac{R^2}{r^2}) = 0$$

$$V_\theta = -V_\infty \sin(\frac{\pi}{2})(1 + \frac{R^2}{r^2}) - \frac{\Gamma}{2\pi r} = 0$$

$${r_s}^2 + \frac{\Gamma}{2\pi V_\infty} r_s + R^2 = 0$$

$$r_s = -\frac{\Gamma}{4\pi V_\infty} \pm \sqrt{(\frac{\Gamma}{4\pi V_\infty})^2 - R^2}$$

When $\frac{\Gamma}{4\pi V_\infty} > R$, $r_s$ results in negative number for all cases. Because both roots are negative, the solution is impossible.

$\underline{Case(2b)}$ : $\theta = -\frac{\pi}{2}$, $r = r_s$

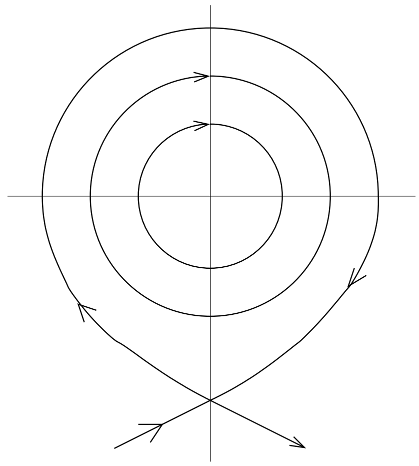
$$V_r = V_\infty \cos(-\frac{\pi}{2})(1 - \frac{R^2}{r^2}) = 0$$

$$V_\theta = -V_\infty \sin(-\frac{\pi}{2})(1 + \frac{R^2}{r^2}) - \frac{\Gamma}{2\pi r} = 0$$

$${r_s}^2 - \frac{\Gamma}{2\pi V_\infty} r_s + R^2 = 0$$

$$r_s = \frac{\Gamma}{4\pi V_\infty} \pm \sqrt{(\frac{\Gamma}{4\pi V_\infty})^2 - R^2}$$

When $\frac{\Gamma}{4\pi V_\infty} > R$, we get $r_s = \frac{\Gamma}{4\pi V_\infty} - \sqrt{(\frac{\Gamma}{4\pi V_\infty})^2 - R^2} < R$. So we can't use this solution.
However, $\theta_s = -\frac{\pi}{2}$ and $\frac{\Gamma}{4\pi V_\infty} > R$ is an acceptable solution when $r_s = \frac{\Gamma}{4\pi V_\infty} \pm \sqrt{(\frac{\Gamma}{4\pi V_\infty})^2 - R^2} > R$



**Force on a Cylinder with Circulation in a Uniform Steady Flow**

Force on an elemental distance on the surface of the cylinder:

$$d\vec{F} = -p_b R d\theta \hat{e_r}$$

$$d\vec{F} = -p_b R d\theta (\cos\theta \hat{i} + \sin\theta \hat{j})$$

$$\vec{F} = \int_0^{2\pi} -p_b R d\theta (\cos\theta \hat{i} + \sin\theta \hat{j})$$

The drag per unit span is

$$D' = \vec{F} \cdot \hat{j} = \int_0^{2\pi} -p_b cos\theta R d\theta$$

The lift per unit span is

$$L' = \vec{F} \cdot \hat{i} = \int_0^{2\pi} -p_b sin\theta R d\theta$$

As we know, in incompressible flow the total pressure $p_o = p + \frac{\rho V^2}{2}$, which is a constant throughout the flow. $p_b = p_o - \frac{\rho(V_r^2 + V_\theta^2)}{2}$. Besides, there is no flow normal to the surface, $V_r = 0$.

$$p_b = p_o - \frac{\rho}{2} V_\theta^2$$

$$p_b = p_o - \frac{\rho}{2}(-2V_\infty \sin\theta - \frac{\Gamma}{2\pi R})^2$$

$$p_b = p_o - 2\rho {V_\infty}^2 (\sin\theta)^2 - \rho V_\infty sin\theta \frac{\Gamma}{R\pi} - \frac{\rho \Gamma^2}{8\pi^2 R^2}$$

$$\therefore D' = R\int_{0}^{2\pi} -(p_o - 2\rho {V_\infty}^2(\sin\theta)^2 - \rho V_\infty sin\theta\frac{\Gamma}{R\pi} - \frac{\rho\Gamma^2}{8\pi^2 R^2})cos\theta d\theta = 0$$

which means that d'Alembert's paradox still prevails.

$$L' = R\int_{0}^{2\pi} -(p_o - 2\rho {V_\infty}^2(\sin\theta)^2 - \rho V_\infty sin\theta\frac{\Gamma}{R\pi} - \frac{\rho\Gamma^2}{8\pi^2 R^2})sin\theta d\theta = \rho V_\infty \Gamma$$

which is the Kutta-Joukowski theorem.

In inviscid, incompressible flow, the resultant force per unit span acting on a 2-D body of any cross section is equal to $\rho V_\infty \Gamma$ and acts perpendicular to $V_\infty$.